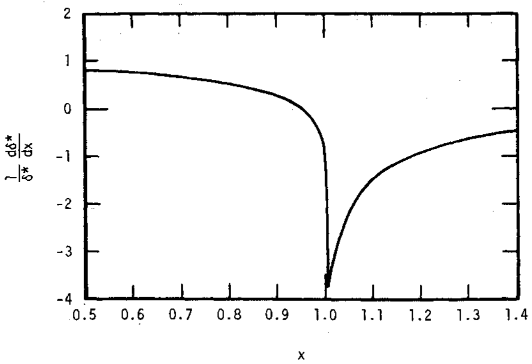

Fig. 1 Relative slopes of the (effective) slender body.

able; however, as illustrated in Fig. 1 the normalized slopes of the (effective) slender body are somewhat larger than 0(1) behind the trailing edge. Thus, the authors decided, within the framework of an "exact" treatment of the over-all problem, to forego linearized airfoil theory.

With respect to omission of cross-streamwise pressure variations in the "Region II" analysis, this was done only after it was determined that resultant errors in calculating the displacement thickness δ^* were no greater than those involved in solving the airfoil problem. (In principle, the "Region I" problem could have been solved with $y \to \infty$; i.e., $t \equiv [1 - \exp(-R^{1/2}y)] \to 1$.) In this respect, the authors reiterate that a primary objective of Ref. 1 was to demonstrate numerical techniques for solving problems of this kind.

References

[1] Schneider, L. I. and Denny, V. E., "Evolution of the Laminar Wake Behind a Flat Plate and Its Upstream Influence," *AIAA Journal*, Vol. 9, No. 4, April 1971, pp. 655–660.

[2] Giesing, J. P., "Solution of the Flow Field about One or More Airfoils of Arbitrary Shape in Uniform and Nonuniform Flows by the Douglas Neumann Method," LB31946, May 1968, Douglas Aircraft Co., Los Angeles, Calif.

Errata

Errata: "Vibrations of Pressurized Orthotropic Shells"

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[AIAA J. **9**, 1201–1203 (1971)]

EQUATIONS (4) should read

$$C_{11}\frac{\partial^2 u}{\partial x^2} + G_{12}\frac{\partial^2 u}{\partial y^2} + (C_{12} + G_{12})\frac{\partial^2 v}{\partial x \partial y} - C_{12}\frac{1}{R}\frac{\partial w}{\partial x} = \rho\frac{\partial^2 u}{\partial t^2} \tag{4a}$$

$$(C_{12} + G_{12})\frac{\partial^2 u}{\partial x \partial y} + G_{12}\frac{\partial^2 v}{\partial x^2} + C_{22}\frac{\partial^2 v}{\partial y^2} - C_{22}\frac{1}{R}\frac{\partial w}{\partial y} = \rho\frac{\partial^2 v}{\partial t^2} \tag{4b}$$

$$-\frac{1}{12}\left(\frac{h}{R}\right)^2\left[C_{11}\frac{\partial^4 w}{\partial x^4} + (2C_{12} + 4G_{12})\frac{\partial^4 w}{\partial x^2 \partial y^2} + C_{22}\frac{\partial^4 w}{\partial y^4}\right] + C_{12}\frac{1}{R}\frac{\partial u}{\partial x} + C_{22}\frac{1}{R}\left(\frac{\partial v}{\partial y} - \frac{w}{R}\right) + \tau_{xx}{}^0\frac{\partial^2 w}{\partial x^2} + \tau_{yy}{}^0\frac{\partial^2 w}{\partial y^2} = \rho\frac{\partial^2 w}{\partial t^2} \tag{4c}$$

Received December 18, 1971.

The second of Eqs. (5) should read

$$v(x, y, t) = B_{mn}\sin(m\pi x/L)\sin(ny/R)\cos\omega t$$

Erratum: "A Nonlinear Method Applied to Variable-Thickness Plates"

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[AIAA J. **10**, 148–154 (1972)]

ON p. 14 , the following heading and paragraph were inadvertently omitted.

Derivations of Equations

The method will be illustrated by solving the large-deflection problem of a rectangular plate of variable thickness under uniform lateral pressure. The nondimensional large-deflection plate equations of von Kármán's, corresponding to compatibility and equilibrium in the lateral direction, can be written as

$$L_f(x, y) = \nabla^4 F - t(w_{,xy}{}^2 - w_{,xx}w_{,yy} - \Omega(y)w_{,xx}) = 0 \tag{1}$$